Accepted Manuscript

Trajectory design for the system of observation of Daytime Asteroids

I.D. Kovalenko, B.M. Shustov, N.A. Eismont

PII: S0094-5765(18)30109-7

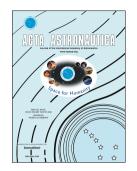
DOI: 10.1016/j.actaastro.2018.05.007

Reference: AA 6863

To appear in: Acta Astronautica

Received Date: 15 January 2018

Revised Date: 29 April 2018 Accepted Date: 2 May 2018



Please cite this article as: I.D. Kovalenko, B.M. Shustov, N.A. Eismont, Trajectory design for the system of observation of Daytime Asteroids, *Acta Astronautica* (2018), doi: 10.1016/j.actaastro.2018.05.007.

This is a PDF file of an unedited manuscript that has been accepted for publication. As a service to our customers we are providing this early version of the manuscript. The manuscript will undergo copyediting, typesetting, and review of the resulting proof before it is published in its final form. Please note that during the production process errors may be discovered which could affect the content, and all legal disclaimers that apply to the journal pertain.

ACCEPTED MANUSCRIPT

Trajectory design for the System of Observation of Daytime Asteroids

I.D. Kovalenko^{a,b,*}, B.M. Shustov^a, N.A. Eismont^b

^aInstitute of Astronomy, Russian Academy of Sciences ^bInstitute of Space Research, Russian Academy of Sciences

Abstract

System of Observation of Daytime Asteroids (SODA) is a space mission project, intended to detect and monitor hazardous asteroids approaching the Earth from the Sun direction, namely from the daytime sky. In particular, the SODA mission aims to detect small celestial objects of size ~ 10 m, approaching the Earth from directions inaccessible to ground-based and near-Earth space-based telescopes, and to inform beforehand whenever a risk of collision with the Earth occurs.

The system uses one or two spacecraft launched into the vicinity of the Lagrangian point L1 of the Sun-Earth system, about 1.5 million km from the Earth. This paper considers main aspects of trajectory design for the SODA project at its initial stage. Selection of an orbit about L1 point, transfer from the Earth and stationkeeping in the vicinity L1 are discussed. A launch scheme making use of a lunar gravity assist is proposed to achieve a desired configuration of the two spacecraft.

Keywords: hazardous asteroid, Sun-Earth Lagrange point L1, lunar gravity assist

Email address: irina.kovalenko@iki.rssi.ru (I.D. Kovalenko)

^{*}Corresponding author

Download English Version:

https://daneshyari.com/en/article/8055500

Download Persian Version:

https://daneshyari.com/article/8055500

<u>Daneshyari.com</u>