### Accepted Manuscript

Influence of different propellant systems on ablation of EPDM insulators in overload state

Yiwen Guan, Jiang Li, Yang Liu, Tuanwei Xu

PII: S0094-5765(17)31306-1

DOI: 10.1016/j.actaastro.2018.01.048

Reference: AA 6678

To appear in: Acta Astronautica

Received Date: 18 September 2017

Revised Date: 1 December 2017

Accepted Date: 23 January 2018

Please cite this article as: Y. Guan, J. Li, Y. Liu, T. Xu, Influence of different propellant systems on ablation of EPDM insulators in overload state, *Acta Astronautica* (2018), doi: 10.1016/ j.actaastro.2018.01.048.

This is a PDF file of an unedited manuscript that has been accepted for publication. As a service to our customers we are providing this early version of the manuscript. The manuscript will undergo copyediting, typesetting, and review of the resulting proof before it is published in its final form. Please note that during the production process errors may be discovered which could affect the content, and all legal disclaimers that apply to the journal pertain.



8

# 1 Influence of Different Propellant Systems on Ablation of EPDM 2 Insulators in Overload State

•	
4	Yiwen Guan <sup>1</sup> , Jiang Li <sup>1,*</sup> , Yang Liu <sup>1</sup> , Tuanwei Xu <sup>2</sup>
5	<sup>1</sup> Science and Technology on Combustion, Internal Flow and Thermal-structure Laboratory, Northwestern
6	Polytechnical University, Xi'an Shaanxi 710072, PR China.

<sup>2</sup>Xi'an Aerospace Power Technology Research Institute, Xi'an Shaani 710025, PR China

\*Corresponding author: Jiang Li; lijiang@nwpu.edu.cn

#### 9 Abstract:

10 This study examines the propellants used in full-scale solid rocket motors (SRM) and investigates how 11 insulator ablation is affected by two propellant formulations (A and B) during flight overload conditions. An 12 experimental study, theoretical analysis, and numerical simulations were performed to discover the intrinsic 13 causes of insulator ablation rates from the perspective of lab-scaled ground-firing tests, the decoupling of 14 thermochemical ablation, and particle erosion. In addition, the difference in propellant composition, and the 15 insulator charring layer microstructure were analyzed. Results reveal that the degree of insulator ablation is positively correlated with the propellant burn rate, particle velocity, and aggregate concentrations during the 16 17 condensed phase. A lower ratio of energetic additive material in the AP oxidizer of the propellant is promising 18 for the reduction in particle size and increase in the burn rate and pressure index. However, the overall higher 19 velocity of a two-phase flow causes severe erosion of the insulation material. While the higher ratio of 20 energetic additive to the AP oxidizer imparts a smaller ablation rate to the insulator (under lab-scale test 21 conditions), the slag deposition problem in the combustion chamber may cause catastrophic consequences for 22 future large full-scale SRM flight experiments.

23

#### 24 Key words:

25 Solid Rocket Motor (SRM); Propellant; Insulator; Ablation; Slag Deposition;

#### 26 1. Introduction

The thermal protection systems of the inner combustion chamber of modern large-scale solid rocket motors (SRM) directly affect the success and reliability of the launch. Owing to the ever-increasing demands in working pressure and combustion temperature, a high-aluminum propellant, containing 17–20% aluminum powder (Al), is widely used, which aims to increase the combustion stability and maximize the specific impulse. However, the use of a high-aluminum propellant, in conjunction with restrictions on the overall size of the SRM, can expose the insulator Download English Version:

## https://daneshyari.com/en/article/8055632

Download Persian Version:

https://daneshyari.com/article/8055632

Daneshyari.com